IN THE CLAIMS:

1. (Currently Amended) A method of coating a gas turbine engine component using a powder coating process to produce a higher melting point coating comprising:

providing a gas turbine engine component having a solid surface;

applying a powder coating to the gas turbine engine component solid surface using the powder coating process, wherein the powder coating is applied in a dry form without an organic solvent, the powder coating comprising a fritted glass matrix with ceramic particles trapped in the matrix, ~~using a fluidized bed or an electrostatic brush~~; and

heating in a single step the applied powder coating to raise the melting point of the glass of the matrix by reacting the ceramic particles with the glass thereby raising the melting point and resulting in a higher engine use temperature, wherein the heating melts and fuses ~~melt and fuse~~ particles of the powder coating to the gas turbine engine component solid surface and ~~cure~~ cures the powder coating.

2. (Currently Amended) The method of claim 1, wherein the powder coating is applied using a fluidized bed process, an electrostatic spray process or an electrostatic brush process.

3. (Canceled)

4. (Currently Amended) The method of claim 2 ~~3~~, wherein the gas turbine engine component is grounded.

5. (Original) The method of claim 4, wherein the powder coating comprises an inorganic based or organic based material.

6. (Currently Amended) The method of claim 5, wherein the powder coating comprises a material ~~is~~ selected from the group consisting of a ~~ceramic,~~ glass/enamel/metal and a composite.

7. (Currently Amended) The method of claim 6, wherein the powder coating comprises a material ~~is~~ selected from the group consisting of silica, alumina, zirconia, magnesium oxide, titanium oxide, yttrium and hafnium oxide.

8. (Original) The method of claim 5, wherein the coating is a thermal barrier coating.

9. (Original) The method of claim 5, wherein the gas turbine engine component is cleaned prior to application of the powder coating.

10. (Canceled)

11. (Currently Amended) The method of claim 1, wherein the component includes an electrically conductive ~~a~~ non-metallic substrate.

12. (Currently Amended) A method of coating a gas turbine engine component using a powder coating process to produce a higher melting point coating comprising:

providing a gas turbine engine component having an electrically conductive solid substrate;

cleaning the gas turbine engine component prior to application of a powder coating;

applying a powder coating to the solid substrate of the gas turbine engine component using the powder coating process, wherein the powder coating is applied in a dry form without an organic solvent, the powder coating comprising a fritted glass matrix with ceramic particles trapped in the matrix; ~~the powder coating process comprising a fluidized bed or an electrostatic brush~~; and

heating in a single step the applied composition to raise the melting point of the glass of the matrix by reacting the ceramic particles with the glass thereby raising the melting point and resulting in a higher engine use temperature, wherein the heating melts and fuses ~~melt and fuse~~ particles of the powder coating to the gas turbine engine component and ~~cure~~ cures the powder coating.

13. (Canceled)

14. (Original) The method of claim 12, wherein heat at a temperature between about 450-1538°C for about 5 minutes to about 24 hours is applied.

15. (Canceled)

16. (Canceled)

17. (Previously Presented) The method of claim 1, wherein the powder coating is applied directly to the gas turbine engine component solid surface or the powder coating is applied to a bond coating located on the gas turbine engine component solid surface.

18. (New) The method of claim 1, wherein a tribo charging process or a corona charging spray process is employed.